

General Disclaimer

One or more of the Following Statements may affect this Document

- This document has been reproduced from the best copy furnished by the organizational source. It is being released in the interest of making available as much information as possible.
- This document may contain data, which exceeds the sheet parameters. It was furnished in this condition by the organizational source and is the best copy available.
- This document may contain tone-on-tone or color graphs, charts and/or pictures, which have been reproduced in black and white.
- This document is paginated as submitted by the original source.
- Portions of this document are not fully legible due to the historical nature of some of the material. However, it is the best reproduction available from the original submission.

**NASA
Technical
Memorandum**

(NASA-TM-86514) IMPROVED TURBOPUMP DYNAMICS
Center Director's Discretionary Fund. Final
Report (NASA) 6 p HC A02/BF A01 CACL 13K

W85-31509

Unclass

G3/37 21815

NASA TM-86514

IMPROVED TURBOPUMP DYNAMICS

**CENTER DIRECTOR'S DISCRETIONARY FUND
FINAL REPORT**

By Larry Kiefling

Systems Dynamics Laboratory
Science and Engineering Directorate

July 1985



National Aeronautics and
Space Administration

George C. Marshall Space Flight Center

1. REPORT NO. NASA TM- 86514		2. GOVERNMENT ACCESSION NO.		3. RECIPIENT'S CATALOG NO.	
4. TITLE AND SUBTITLE Improved Turbopump Dynamics, Center Director's Discretionary Fund Final Report				5. REPORT DATE July 1985	
				6. PERFORMING ORGANIZATION CODE	
7. AUTHOR(S) Larry Kiefling				8. PERFORMING ORGANIZATION REPORT #	
9. PERFORMING ORGANIZATION NAME AND ADDRESS George C. Marshall Space Flight Center Marshall Space Flight Center, Alabama 35812				10. WORK UNIT NO.	
				11. CONTRACT OR GRANT NO.	
12. SPONSORING AGENCY NAME AND ADDRESS National Aeronautics and Space Administration Washington, D.C. 20546				13. TYPE OF REPORT & PERIOD COVERED Technical Memorandum	
				14. SPONSORING AGENCY CODE	
15. SUPPLEMENTARY NOTES Prepared by Systems Dynamics Laboratory, Science and Engineering Directorate.					
16. ABSTRACT A study was initiated to investigate the practicality of increasing rotor critical speeds by changes in manufacturing method. The technique would be to build a pump with an all-laser-welded shaft and case; such unit to be opened by laser cutting and rebuilt by rewelding the same surface. Use of a split casing, common in industry, would permit assembly of the rotor outside the case. A team was formed to perform the study; however, the work of the team was severely restricted by conflict with higher priority tasks. No manpower was available to evaluate alternate configurations. Thus, much of the synergetic effects of cohesive design modification was lost. Although very limited results were achieved, nothing was found to indicate that the method is not worth further investigation.					
17. KEY WORDS Turbopump Dynamics			18. DISTRIBUTION STATEMENT Unclassified - Unlimited		
19. SECURITY CLASSIF. (of this report) Unclassified		20. SECURITY CLASSIF. (of this page) Unclassified		21. NO. OF PAGES 5	
				22. PRICE NTIS	

TECHNICAL MEMORANDUM

IMPROVED TURBOPUMP DYNAMICS

CENTER DIRECTOR'S DISCRETIONARY FUND FINAL REPORT

INTRODUCTION

Critical speed limitations are among the most serious design restraints for state-of-the-art rocket engine turbomachinery. Increases in critical speeds are highly desirable. Increases in critical speed can be achieved by making the rotor shorter and more rigid.

APPROACH

Establish feasibility of building a pump with an all-laser-welded shaft and case; such unit to be opened by laser cutting and rebuilt by rewelding the same surface. Use of a split casing, common in industry, would permit assembly of the rotor outside the case. A sacrificial strip would be used to limit cutting damage, and each surface would have protected precision locating surfaces (Fig. 1). Establish a revised pump preliminary design using the existing major fluid handling components but optimizing dynamic characteristics and other components. Determine the performance increases possible in the following areas:

1. Primary

- a. Increase or rotor critical speed
- b. Decrease in rotor damping
- c. Improved balance

2. Secondary

- a. Seal performance improvement
- b. Weight reduction
- c. Performance improvement
- d. Cost reduction (reduced number of machined parts and surfaces).

A team was formed to conduct the study. The SSME HPOTP (high pressure oxygen turbopump) was selected as the target unit.

RESULTS

The work of the team was severely restricted by conflict with higher priority tasks. No manpower was available to evaluate alternate configurations. Thus, much of the synergetic effects of cohesive design modification was lost.

Structural dynamic modes were calculated for a welded rotor versus the present built-up rotor (Fig. 2) with the same length and external dimensions. A frequency increase of 5 percent was calculated. An increase of this magnitude is not considered significant.

Electron beam welding has been identified as possibly more desirable than laser welding for the applications and a subject for further evaluation.

Further research tasks have been identified to support further development and address key problems. These consist of coupon tests to evaluate the cut and reweld concept and development of distortion minimization techniques.

CONCLUSION

The goals of this study proved to be beyond the resources of a small team working on a part time basis. Accomplishment will require a diversified, dedicated design team working under guidelines to allow major changes from present design. Implementation of the welding-cutting technique on a piece-wise basis is not practical.

Key problems with the technique have been identified and are being made subject to further research.

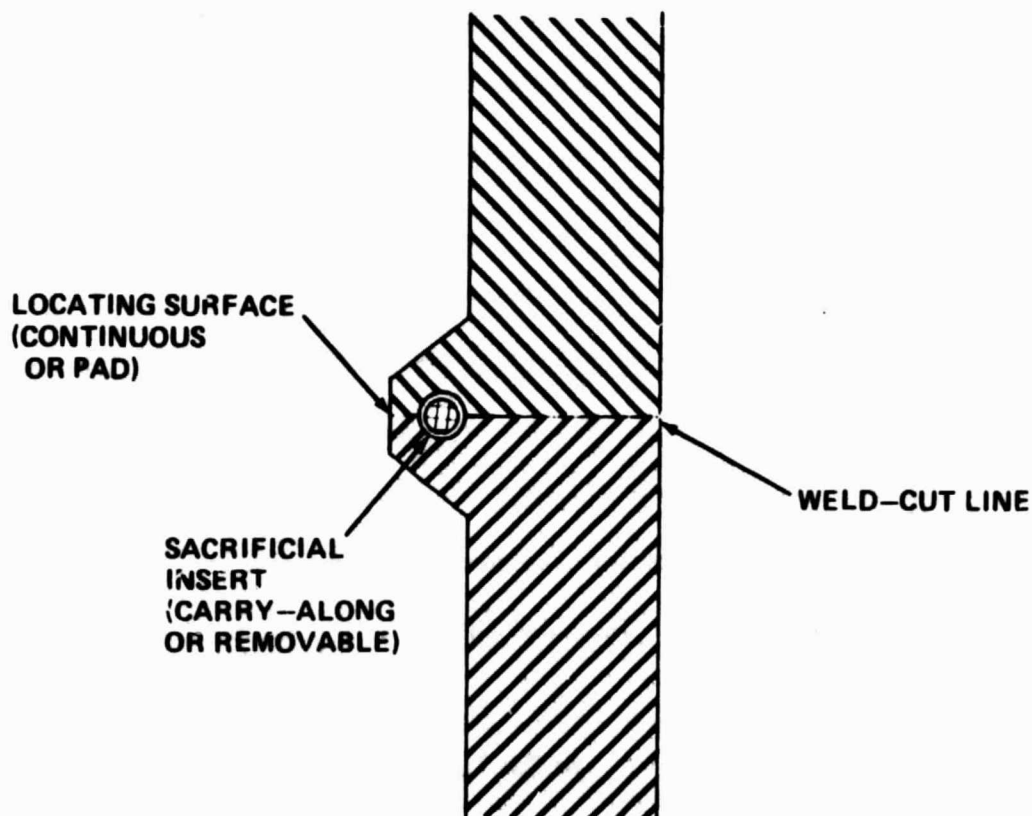


Figure 1. Proposed joint configuration pump construction.

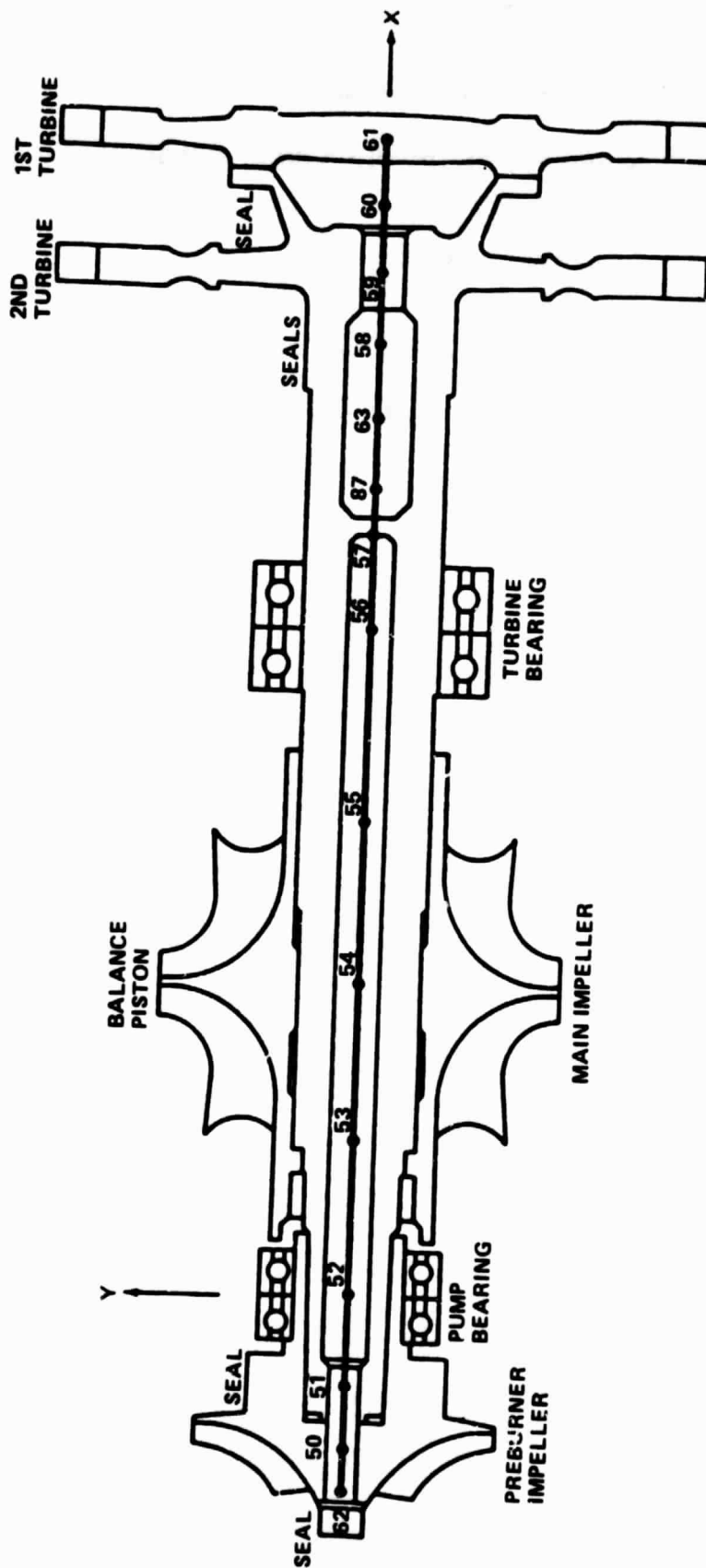


Figure 2. SSME HPOTP rotor and joints on current math model.

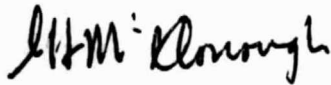
APPROVAL

IMPROVED TURBOPUMP DYNAMICS

CENTER DIRECTOR'S DISCRETIONARY FUND FINAL REPORT

By Larry Kiefling

The information in this report has been reviewed for technical content. Review of any information concerning Department of Defense or nuclear energy activities or programs has been made by the MSFC Security Classification Officer. This report, in its entirety, has been determined to be unclassified.



G. F. McDONOUGH
Director Systems Dynamics Laboratory